

ATA AIRLINES, INC.

T/E FLAP LOAD RELIEF SYSTEM

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CHECK BEING PERFORMED: Custom

ZONES: 119 212
A/C NUMBER:
REV. DATE: 04/21/99
FREQUENCY: 1C

W/C NUMBER: 221M2705 DATE:
W/O:
JAC CODE:

PANELS

119BL

REFERENCES

1

MECH INSP

OPERATIONALLY CHECK T/E FLAP LOAD RELIEF SYSTEM.

1 T/E Flap Load Relief - System Test

A References

- 1 06-41-00/201, Fuselage Access Doors and Panels
- 2 24-22-00/201, Electrical Power - Control
- 3 29-11-00/201, Main (Left, Right, Center)
Hydraulic Systems
- 4 32-00-15/201, Landing Gear Door Locks

B Access

1 Location Zones

- 143 Left MLG Wheel Well
- 197/198 Wing to Body - Aft Lower Half (Left/Right)
- 211/212 Control Cabin - Section 41 (Left/Right)

C Prepare for the Test

- _____ XXXXX 1 Supply electrical power (Ref 24-22-00/201).
- _____ XXXXX 2 Put the EICAS computer switch on the EICAS DISPLAY
select panel, P9, to L.
- _____ XXXXX 3 Make sure the six EICAS circuit breakers on the
overhead panel, P11, are closed.
- _____ XXXXX 4 Make sure the flaps and slats are in the fully
retracted position with the flap control lever in the

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ATA AIRLINES, INC. B757 FLEET

W/C #: 221M2705

DATE WORK CARD COMPLETE ___/___/___

zero (FLAPS UP) detent.

WARNING: KEEP PERSONS AND EQUIPMENT AWAY FROM ALL CONTROL SURFACES WHEN HYDRAULIC POWER IS SUPPLIED. AILERONS, ELEVATORS, RUDDER, FLAPS, SLATS, SPOILERS, AND STABILIZER ARE FULLY POWERED SURFACES. INJURY TO A PERSON OR DAMAGE TO EQUIPMENT CAN OCCUR WHEN HYDRAULIC POWER IS SUPPLIED.

_____ XXXXX 5 Pressurize the left hydraulic system (Ref 29-11-00/201).

D T/E-Flap Load Relief System - Test

_____ XXXXX 1 Move the flap control lever to the 30-unit detent, and let the T/E flaps and L/E slats fully extend.

_____ XXXXX 2 Open this circuit breaker on the P11 panel and attach DO-NOT-CLOSE tag:

a 11J18, FLAP LOAD RELIEF

_____ XXXXX 3 Push and hold the LOAD RELIEF--test switch.

NOTE: The LOAD RELIEF--test switch is near the E/E bay access panel.

_____ XXXXX 4 Make sure the EICAS message, FLAP LD RELIEF, does show on the display unit and the TRAILING EDGE light is on.

_____ XXXXX 5 Release the LOAD RELIEF--test switch.

_____ XXXXX 6 Make sure the EICAS message, FLAP LD RELIEF, does not show on the display unit and the TRAILING EDGE light is off.

_____ XXXXX 7 Remove the DO-NOT-CLOSE tag and close this circuit breaker on the P11 panel:

a 11J18, FLAP LOAD RELIEF

_____ XXXXX 8 Push and hold the LOAD RELIEF--test switch and do the checks that follow:

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CHECK BEING PERFORMED: Cust

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MECH: INSP:

- a Make sure the T/E flaps move to the 25-unit position.
- b Make sure the L and R needles, on the flap position indicator, are in the tolerance band for the 25-unit position.

_____ XXXXX 9 Release the LOAD RELIEF--test switch and do the checks that follow:

- a Make sure the T/E flaps go to the 30-unit position.
- b Make sure the position indicator needles are in the tolerance band for the 30-unit position.

_____ XXXXX 10 Make sure the EICAS messages, FLAP/SLAT ELEC or FLAP ISLN VAL, do not show on the status display.

_____ XXXXX 11 Make sure the message, FLAP/SLAT ELEC, FLAP ISLN VAL, or a FLAP/SLAT BITE, do not show on the ECS/MSG display.

_____ XXXXX 12 Move the flap control lever to the zero (FLAPS UP) detent, and let the T/E flaps and L/E slats fully retract.

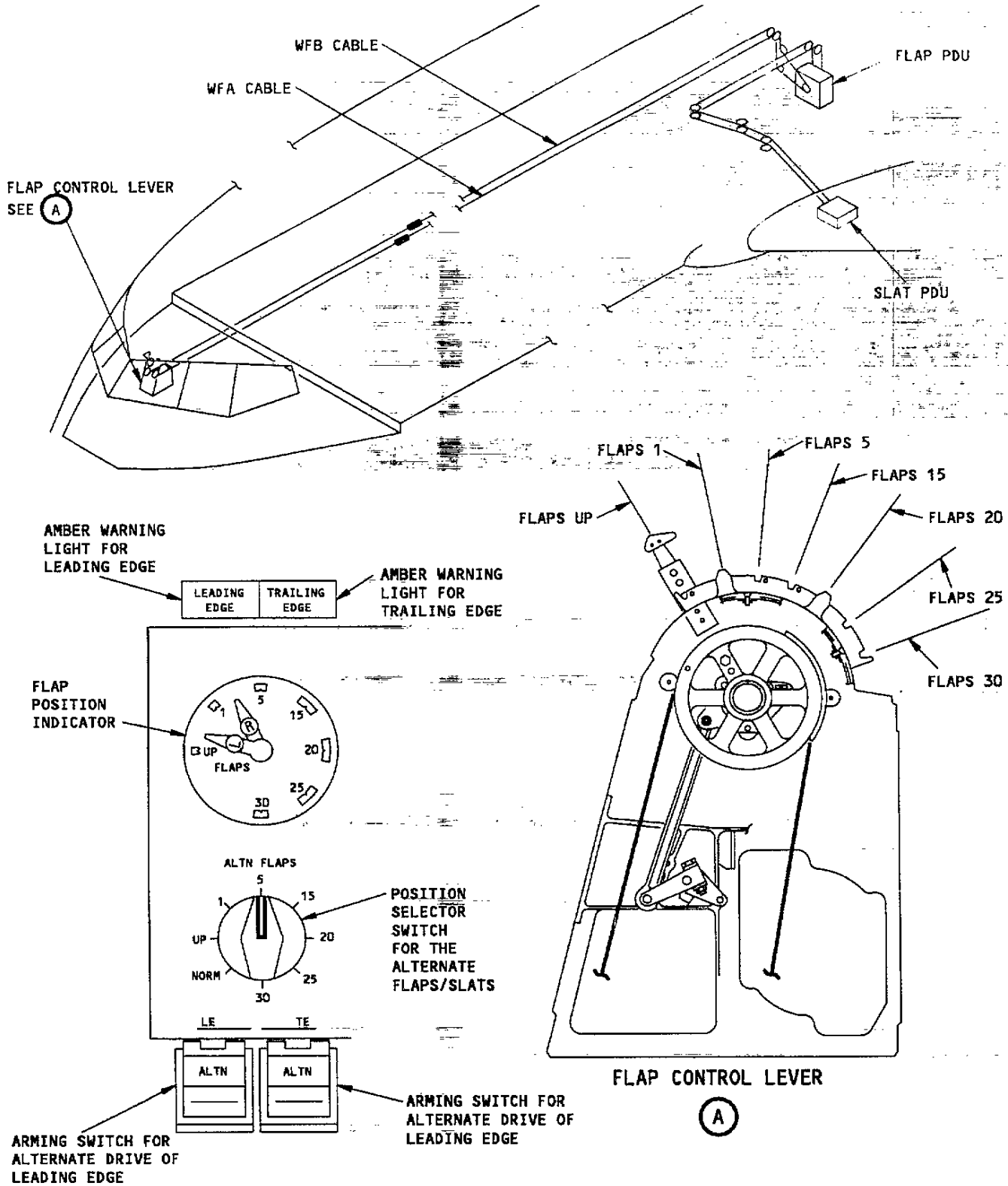
E Put the Airplane Back to Its Usual Condition

_____ XXXXX 1 Remove the power from the left hydraulic system (Ref 29-11-00/201).

_____ XXXXX 2 Remove the electrical power if it is not necessary (Ref 24-22-00/201).

*****END OF WORKCARD*****

221M2705



Trailing Edge Flap Controls